

ABSTRACT

The invention relates to an aircraft engine, particularly a gas turbine engine. The aircraft engine comprises at least one fan (11) and a core engine (12). The fan (11) comprises a fan housing (13) which encloses a fan flow channel and at least one fan wheel (15). The core engine (12) comprises at least one compressor (15, 16), at least one combustion chamber (17) and at least one turbine (18, 19). The aircraft engine further comprises at least one generator (24) for producing electrical energy whereby the or each generator (24) produces electrical energy by withdrawing shaft power from the core engine (12). According to the invention the or each generator (24) for producing electrical energy is integrated into at least one strut (21) extending in the radial direction of the fan flow channel and is thus positioned within the fan flow channel.